



SCHWEIZER SERVICE BULLETIN

B-265.1*
23 Jul 1996

MANDATORY

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SUBJECT: 269A6050-005 LTS TAIL ROTOR PITCH CONTROL ASSEMBLY, TORQUE VERIFICATION OF 269A6258 NUT, AND APPLICATION OF TORQUE STRIPES

MODELS AFFECTED: All Model 269A, TH-55A, 269A-1 and 269B Helicopters, and 269C Helicopters up to and including Serial Number S1731

- Any Model 269A, TH-55A, 269A-1, 269B or 269C Helicopter equipped with a 269A6050-005 pitch control assembly having a Serial Number prior to S0942
- Any spare 269A6050-005 pitch control assembly having a Serial Number prior to S0942
- Pitch control assemblies with the following Serial Numbers were in compliance with this Service Bulletin at delivery and require no further action: S0714, S0793, S0802, S0839, S0875, S0895, S0900, S0904, S0905, S0907, S0910, S0914 through S0918, S0920 through S0922, S0924, S0925, S0927 through S0933, S0935 through S0938 and S0941
- Pitch control assemblies that have been disassembled and reassembled in the field in accordance with HMI Appendix C, Part VII, Section 9 (revised 07 Nov 1995, or later revision) are already in compliance with this Service Bulletin and require no further action

TIME OF COMPLIANCE:

- For affected 269A6050-005 pitch control assemblies with Serial Numbers S0783 through S0941; perform PART I within next 25 hours of operation
- For all affected 269A6050-005 pitch control assemblies: perform PART II at next scheduled 100-Hour/Annual Inspection
- For all affected spares: perform PART II prior to installation on helicopter or within one year from date of this Service Bulletin, whichever occurs first

REFERENCE: Basic HMI (Issued: 15 Mar 1982; Revised 19 Feb 1996)

HMI Appendix C, PART VII, Section 9 (Issued: 15 Mar 1976; Revised 07 Nov 1995, or later revision)

PREFACE: ● Reports indicate that rotational slippage of the swashplate and loosening of the retention nut has occurred

- Failure to comply with this Service Bulletin may lead to loss of control of the helicopter, and subsequent serious injury, death and/or property damage.

(■) Denotes portion of text added or revised.

*Supersedes B-265, dated 25 Sep 1995

1 of 2

PARTS, TOOLS AND MATERIAL REQUIRED

PART NUMBER	NOMENCLATURE	SOURCE	QTY
269A6051	Lockwasher	SAC	1
HS1551S290	Tail Rotor Attachment Washer	SAC	1
369A9822-3	Pitch Control Assembly Holding Block	SAC	1
269A9307/269T9307	Locknut Torque Wrench Adapter	SAC	1
269T9275-3	Tail Rotor Wrench	SAC	1
---	White Lacquer Paint	Commercial	A/R

PROCEDURE, PART I:

- a. Cut lockwire and retract boot to gain access to pitch control assembly. Using firm hand pressure, check for looseness of 269A6258 nut.
- b. If hand check reveals looseness of 269A6258 nut, perform PART II prior to next flight.
- c. If hand check appears to be satisfactory (no looseness) perform PART II as specified under TIME OF COMPLIANCE.

PROCEDURE, PART II:

- a. Remove tail rotor and pitch control assembly from helicopter (Basic HMI Section 9). (Match mark assembly to aid in reinstallation.)

NOTE

Perform steps b. through g. in accordance with HMI Appendix C, Part VII, Section 9

- b. Disassemble tail rotor pitch control from tail rotor assembly.
- c. Remove 269A6258 nut; discard 269A6051 lockwasher.
- d. Inspect 269A6049-3 shaft to bearing joint for signs of fretting, slippage or damage; also check for secure fit. If necessary, remove 269A6049-3 shaft and inspect.
- e. Reassemble shaft to bearing as required.
- f. Install new 269A6051 lockwasher.
- g. Install 269A6258 nut and torque; apply torque stripes.
- h. Install pitch control and tail rotor assembly on helicopter (Basic HMI Section 9).
- i. Record compliance with this Service Bulletin in the component and aircraft records.
- j. Notify SAC Customer Service if rotational slippage or loosening of the retention nut has occurred.

WEIGHT AND BALANCE

Weight and balance are not affected.