



SCHWEIZER SERVICE NOTICE

NOTICE NO. N-222

DATE: Sept. 21, 1988

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MANDATORY

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SUBJECT: TEXTRON LYCOMING SERVICE BULLETINS:
o No. 480. Oil and Filter Change and Cleaning.
o No. 369G. Engine Inspection after Overspeed or Overboost.

MODELS AFFECTED: As listed on subject Service Bulletins.

TIME OF COMPLIANCE: As required by Subject Service Bulletins.

REFERENCE: TEXTRON Lycoming Service Bulletin No. 480, Dated January 18, 1988.
TEXTRON Lycoming Service Bulletin No. 369G, Dated September 7, 1987.

PREFACE: Your attention is directed to the subject TEXTRON Lycoming Service Bulletins No. 480 and 369G, reprinted as part of this Service Information Notice.

These bulletins provide requirements for changing oil and oil filter and for inspecting aircraft engines following engine overspeed or overboost. Owners and Operators of 269 Series Helicopters equipped with an affected model engine, incorporating a pressure screen system, should contact their local Lycoming Dealer to obtain an approved oil filter replacement kit. This kit will increase the 25-Hour interval oil change to a 50-Hour interval. (Also refer to HMI Appendix B, Table B-3 for other required inspections after engine speed exceeds 2000 rpm with the rotors disengaged.) Schweizer Aircraft Corp. recommends that compliance with the subject bulletins be accomplished as specified within the bulletins.

It should be noted that Owners and Operators of 269 Series Helicopters should maintain a complete set of TEXTRON Lycoming Service Letters, Bulletins, and Instructions (Lycoming Publication No. BS-148) as part of the primary information file for the helicopter. (Refer to Table 2-7 of Basic HMI for other related publications and directives.)

TEXTRON Lycoming

Williamsport Plant
Textron Lycoming/Subsidiary of Textron Inc.
652 Oliver Street
Williamsport, PA 17701 U.S.A.

SERVICE BULLETIN

DATE: January 18, 1988 Service Bulletin No. 480
Engineering Aspects are
FAA Approved

SUBJECT: Oil and Filter Change and Cleaning

MODELS AFFECTED: All Textron Lycoming Direct Drive and TIGO-541 Piston Engines.

TIME OF COMPLIANCE: As required by subject bulletin.

Textron Lycoming recommends the following:

- A. 50-Hour interval oil change and filter replacement for all engines using full-flow filtration system.
- B. 25-Hour interval oil change and screen cleaning for all engines employing a pressure screen system.
- C. A total of four (4) months maximum between changes for both systems listed under "A" and "B".
- D. The use of ashless dispersant oil (MIL-L-22851 specifications) is recommended by Textron Lycoming for all new, newly overhauled and re-manufactured Turbocharged Engines after the initial break-in period of 25-50 hours. The exception to this are the following models: TO-360C & F, TIO-360C and TIO & TIGO-541, which must use ashless dispersant oil from the start.

This service bulletin supersedes all previously published recommendations.

TEXTRON Lycoming

Williamsport Plant
Textron Lycoming/Subsidiary of Textron Inc.
652 Oliver Street
Williamsport, PA 17701 U.S.A.

SERVICE BULLETIN

DATE: September 7, 1987

Service Bulletin No. 369G
(Supersedes Service Bulletin No. 369F)
Engineering Aspects are
FAA Approved

SUBJECT: Engine Inspection After Overspeed or Overboost

MODELS AFFECTED: All Avco Lycoming piston engines.

TIME OF COMPLIANCE: As required by the subject bulletin.

PART I - OVERSPEED

As shown in Chart I, every Avco Lycoming piston engine is rated at a specified rpm value above which it may not be operated safely. Operating above the rated engine speed can accelerate wear of stressed parts, possibly resulting in their damage or failure. Momentary overspeed can occur during a landing attempt, when the prop governor lags as the throttle is suddenly opened for a go-around.

For fixed-wing aircraft, momentary overspeed is defined as an increase of no more than 10% of rated engine RPM for a period not exceeding 3 seconds. For rotary-wing aircraft, overspeed is defined as operating at any speed above rated engine RPM for any period of time. No momentary overspeed is allowed for rotary-wing aircraft.

CAUTION

ENGINES MAY NOT BE CONTINUOUSLY OPERATED ABOVE SPECIFIED MAXIMUM CONTINUOUS RPM; TO DO SO WILL RESULT IN ABNORMAL WEAR ON BEARINGS, COUNTERWEIGHT ROLLERS AND OTHER ENGINE PARTS, CONCLUDING IN EVENTUAL ENGINE FAILURE.

Because inadvertent overspeed does occur, the information in this Service Bulletin is provided as an inspection procedure for an engine subjected to overspeed. Record any instance of overspeed in the engine log, along with the corrective action taken. Also note that the engine was inspected per this Service Bulletin.

NOTE

A few models have a 5 minute take-off rating in addition to the continuous rating. On these engines, if overspeed does not exceed the take-off rating for longer than five minutes it may be disregarded. Also, for these engines the take-off rating may be considered to be the maximum rated speed when considering any momentary event of overspeed. Some engines, even though possessing parts of the same structural integrity, have different HP & RPM ratings. In these cases, when computing overspeed, the greater RPM may be used. (Reference Chart I.)

After locating the rated speed in Chart I, find the column for it in Chart II; then determine the percentage of overspeed from the values shown in the applicable column. For example, if the rated speed of the engine was found to be 2800 rpm and the overspeed was 2900 rpm, then from the 2800 column it can be determined that percentage of overspeed is less than 5%. In the lower portion of the chart, across from "2800" and down from 5% or less, locate the number "1" indicating that the instructions in Note 1 should be followed before the engine is returned to service.

CAUTION

IF OVERSPEED EXCEEDS 10% OF THE RPM VALUES IN THE COMPUTING-OVERSPEED COLUMN IN CHART I, IT IS RECOMMENDED THAT THE PROPELLER MANUFACTURER BE CONTACTED FOR POSSIBLE PROPELLER INSPECTION PROCEDURES.

NOTES

1. Determine the cause for overspeed and correct it.
2. Drain the lubricating system.
 - a. Remove oil screens and filters and inspect for metal contamination.
 - b. Perform a differential pressure check on all cylinders to determine the sealing quality of the rings and valves. See Service Instruction No. 1191 for procedure.
 - c. Using a borescope or equivalent instrument, examine the walls of each cylinder for scoring, which could be caused by stuck or broken piston rings.
 - d. Disassemble magnetos and inspect all components for damage; recondition or replace parts as required. Reassemble and test in accordance with the applicable magneto overhaul instruction manual. Also inspect condition of the magneto drive gears on the engine for looseness, which would indicate the supporting idler shafts are loose due to failure of safety attachments. If applicable, inspect condition of magneto bearing recess in crankcase for excessive wear. Repair as necessary in accordance with Service Instruction No. 1140 or No. 1197. (Slick magneto series 4000 and 4100 are not repairable and should be replaced. Slick 4200 series are repairable.)
3. On mechanically supercharged engines, remove the supercharger drain cover and look for presence of engine lubricating oil which, if found, is indicative of a damaged supercharger seal. To determine the extent of damage, permit the oil to drain from the supercharger for a period of 8 hours; if the quantity of oil accumulated is more than a teaspoonful, the supercharger seal should be replaced.
4. Disconnect both the inlet and outlet attaching hardware from the turbocharger and examine the compressor and turbine wheels for possible damage. Check the shaft-wheel assembly for free turning and for vertical and lateral motion, which is indicative of damaged center housing bearings. Damage in these areas must be corrected before the engine is returned to service.
5. Either repeated moments or short periods of operation in the overspeed region accelerate the rate of wear in the parts that comprise the valve train and consequently reduce the reliability of the engine. In addition to the checks normally performed on the engine during a 100 hour periodic maintenance inspection, also accomplish the following steps on page 5 before the aircraft is returned to service: -

Chart I - Specified Rated Engine RPM

Engine Models	SPECIFIED ENGINE SPEED		
	Continuous Rated RPM	5 Minute Take-Off Rating	RPM For Computing Overspeed
O-235-C1, -C1B, -C2A, -C2B, -E, -F, -G, -J; -K2A, -L2A, -M, -N, -P -C1C, -H2C, -L2C, -C1C, -K2C	2800		2800
O-290-D, -D2	2600	2800	2800
O-320-A, -B, -C, -D, -E, -H; IO-320; LIO-320-B, -C; *AIO-320-A, -B, -C; *AEIO-320-E	2700		2700
O-320-E2A, -E2C, -E2F; *AEIO-320-E2A (rated at 140 hp)	2450		2700
O-340-A, -B	2700		2700
O-360-A, -B, -C (except -C2D), -D, -E, -F, -G; IO/LIO-360-C, LO-360-E; *AIO-360; VO-360; IVO-360, HO-360-A1A; *AEIO-360	2700		2700
O-360-C2D	2700	2900	2900
HO-360-B; HIO-360-A, -B, -C, -E	2900		2900
HIO-360-D1A	3200		3200
HIO-360-F1AD	3050		3050
TO-360-A, -C, -E, -F; LTO-360-A; TIO-360-A, -C	2575		2575
O-435-A, -C	2550		2550
GO-435-C2	3100	3400	3400
VO-435-A	3200	3400	3400
VO-435-B; TVO-435	3200		3200
GO-480-B	3000	3400	3400
GO-480-C, -D, -F, -G, -H; IGO-480	3100	3400	3400
GSO-480; IGSO-480	3200	3400	3400
O-540-A, -B, -D; IO-540-C, -J	2575		2700
IO-540-A, -B, -E, -G, -P; HIO-540-A; TIO/LTIO-540 (except -S, -V)	2575		2575
O-540-J, -L; IO-540-W	2400		2400
IO-540-AA1A5	2425		2425
O-540-E, -G, -H; IO-540-D, -K, -L, -M, -N, -R, -S, -T, -V; TIO-540-S	2700		2700
TIO-540-V	2600		2600
AEIO-540-D, -L	2700		2700
VO-540-A	3200	3300	3300
VO-540-B; IVO-540; TIVO-540; VO-540-C	3200		3200
IGO-540	3000	3400	3400
IGSO-540	3200	3400	3400
TIO-541-A1A	2575		2575
TIO-541-E	2900		2900
TIGO-541	3200		3200
IO-720 (400 hp)	2650		2650
IO-720-D1CD (375 hp)	2400		2650

* Aerobatic engines that are engaged in flight maneuvers which cause engine overspeed are subject to abnormal wear and possible overstress of rotating parts, which will shorten the service life of the engine. The damage accumulated due to the amount of overspeed, along with the extent of repeated operation at alternating high and low power applications, must be evaluated by the operator to determine the inspection procedures required.

Chart II - Inspection Requirements in Event of Overspeed

Engine Overspeed in Excess of Max. Rated RPM	ENGINE RPM												
	2400	2425	2550	2575	2600	2650	2700	2800	2900	3050	3200	3300	3400
* 5%	2520	2546	2675	2705	2730	2780	2835	2940	3045	3202	3360	3465	3570
*10%	2640	2668	2800	2830	2860	2915	2970	3080	3190	3355	3520	3630	3740

* Except as defined as "Momentary Overspeed" in 2nd paragraph of Service Bulletin.

Category of Engine Types and Inspection Requirements
(Numbers Refer to Notes in Body of Text)

Specified Engine Speed	FIXED WING INSTALLATIONS									ROTARY WING INSTALLATIONS		
	DIRECT DRIVE (Normally Aspirated)			DIRECT DRIVE (Turbocharged)			GEARED DRIVE			5% or less	between 5-10%	over 10%
	5% or less	between 5-10%	over 10%	5% or less	between 5-10%	over 10%	5% or less	between 5-10%	over 10%			
2400	1	1,2,5	1,6									
2425	1	1,2,5	1,6									
2550	1	1,2,5	1,6									
2575	1	1,2,5	1,6	1	1,2,4,5	6				1,2	1,2,5	6
2600				1	1,2,4,5	6						
2650	1	1,2,5	1,6									
2700	1	1,2,5	1,6							1,2	1,2,5	6
2800	1	1,2,5	1,6									
2900				1	1,2,4,5	6				1,2	1,2,5	6
3050										1,2	1,2,5	6
3200							1,3	1,3,4,5	6	1,2	1,2,5	6
3300										1	2,5	6
3400							1,3	1,3,4,5	6			

NOTES (continued)

- a. Inspect all screens and filters in the lubrication system for metal contamination; if any unexplainable accumulation is discovered, the cause must be determined and corrected before the engine is returned to service.
 - b. By means of a borescope or equivalent illuminated magnifying optical device, determine the condition of the intake and exhaust valve faces and seat faces. Evidence of excessive wear, pounding, or grooving is reason for the valve and seat replacement.
 - c. Inspect external condition of valve keys, rockers, and exhaust valve guides for damage; particularly check valve springs for coil strikes or severe bottoming of the coils. If damage to springs is evident, remove them and check compression load as specified in Table of Limits; replace any that are not within limits.
 - d. Rotate the crankshaft by hand to determine if valve lift is uniform or equal for all cylinders; also note if valve rockers are free when the valves are closed. Unequal valve lift is an indication of bent push rods; and tight rockers, when valves are closed, indicate a tuliped valve or a damaged valve lifter. Repair any suspected damage before the engine is returned to service.
 - e. Comply with Service Bulletin No. 388 to determine exhaust valve stem to valve guide clearance condition.
6. Remove the engine from the aircraft; disassemble it and inspect the parts in accordance with the applicable overhaul manual. Replace any parts that are damaged or not within the service limits as shown in the Table of Limits. In engines equipped with dynamic counterweights, the bushings must be replaced in both counterweight and crankshaft.

PART II - OVERBOOST

The maximum manifold pressure of turbocharged engines is controlled by various means:

1. Throttle controlled by the pilot. Here maximum rated manifold pressure is red-line and is normally reached somewhere before full-open throttle, depending upon density altitude.
2. Preset density controller. This controller senses compressor discharge density and varies the manifold pressure to ensure the engine develops rated power, up to critical altitude, regardless of the density altitude. Here takeoff is at full throttle. However, the red line on the manifold pressure gage is the maximum permissible for a hot day at high field elevation. See the airframe or engine operator's manual for standard day manifold pressure, realizing that full rated power will require a lower manifold pressure on a below standard temperature day and higher on an above standard day. This further indicates that should density controller be improperly adjusted or malfunction, it is possible to have an overboost without exceeding red-line manifold pressure.
3. Preset absolute variable pressure controller. This controller is normally used on engines incorporating a turbo compressor air bleed to pressurize the aircraft cabin. The controller is preset at the factory resulting in red-line manifold pressure at full throttle regardless of density altitude.

Overboost of Avco Lycoming supercharged or turbocharged engines is not permitted beyond the limiting manifold pressure which appears on the Sea Level and Altitude Curves of the applicable Avco Lycoming Operator's Manual. Any operation of an engine beyond this limit raises the possibility of serious engine damage. Because of this, any overboost, whether malfunction or inadvertent, which exceeds the allowable

manifold pressure specified for the corresponding ambient pressure and temperature should be considered as shown in the following chart. It is the responsibility of the operator to monitor manifold pressure to ensure limits are not exceeded. The continued use of an engine after momentary overboost has occurred is at the discretion and the responsibility of the operator.

NOTE

During takeoff with low oil temperature, advancing the throttle too quickly may result in manifold pressure "overshoot". What happens is that manifold pressure advances momentarily above maximum rated by 1 or 2 inches Hg. and then returns immediately to the maximum rated. In this instance, the throttle is slightly ahead of the controller's capacity to function normally. If overshoot does not exceed 2 inches and 3 seconds duration, it may be disregarded. However, overshoot can be prevented by interrupting the throttle advance momentarily several inches below rated manifold pressure.

Chart III - Overboost Recommendations (No Manifold Pressure Relief Valve)

IN ALL CONDITIONS LISTED BELOW, INVESTIGATE REASON FOR OVERBOOST
AND CORRECT BEFORE NEXT FLIGHT

Overboost Conditions	Recommendations
Momentary overboost not exceeding 3 inches Hg. for 5 seconds.	Enter in log book. Include maximum manifold pressure reached, duration of overboost, cylinder head temperature, ambient air temperature, pressure altitude.
Not exceeding 5 inches Hg. or 10 seconds.	Normal 50 hour inspection plus particular attention to items 1, 2 and 3 in the following list.
Not exceeding 10 inches Hg.	Remove engine from aircraft; completely disassemble and inspect. Replace all parts that do not come within maximum service limits as shown in Avco Lycoming Service Table of Limits.
Over 10 inches Hg.	Complete engine overhaul required, plus replacement of crankshaft

1. Inspect cylinder assemblies for signs of cracked heads, particularly around the lower spark plug holes; and for cracks around the hold-down flange of cylinder barrels. Also check barrels for burned paint and for oil leaks around cylinder base flanges.
2. Remove oil screens and inspect for metal particles using care to insure the particles are metal and not hard carbon.
3. Remove all spark plugs and inspect them closely for physical and structural defects. Spark plugs removed may be reused providing that each plug checks out satisfactorily in a spark plug test unit and exhibits none of the following defects:
 - a. Fine wire plugs with loose center or ground electrodes.
 - b. Electrodes show signs of metal or impact damage.

- c. Massive electrode plugs with copper run-out of center electrode.
- d. Ceramic core nose with a cracked or crazed surface.

PART III - ENGINES EQUIPPED WITH ABSOLUTE PRESSURE RELIEF VALVE

Some Avco Lycoming turbocharged engines are equipped with an absolute pressure relief valve (often referred to as a "pop-off" valve). This valve is installed between the compressor outlet and the fuel injector/carburetor to protect the engine from surges of excessive manifold pressure. Even though manifold pressure may continue to rise above its normal rated value, power output will not increase appreciably. In fact, as the valve lifts off its seat, at approximately 2 inches above normal-rated, power may decrease even if manifold pressure continues to rise above normal-rated pressure.

Action required with manifold pressure overboost:

1. Determine the cause for overboost and correct it.
2. Remove the absolute pressure relief valve (pop-off valve).
3. Place the valve assembly, mounting flange down, on a calibrated scale. The valve head should protrude approximately 0.02 inch below the mounting flange. Ensuring that the mounting flange remains parallel to the scale surface, apply pressure to the top of the valve housing. If the valve head depresses flush with the mounting flange surface, without exceeding the maximum pounds of force listed below, the valve is functioning.

In the following chart, pressure relief valves are divided into three categories, according to manifold pressure requirements.

Categories - Manifold Pressure Required to Develop Rated Power	Maximum Pounds To Depress Valve
30.00 to 40.00 inches Hg.	* 43 lbs.
40.00 to 50.00 inches Hg.	* 58 lbs.
50.25 to 60.00 inches Hg.	* 72 lbs.

* Any lesser pressure is acceptable as long as the valve, when in service, does not lift off its seat prior to attaining rated manifold pressure.

NOTE

After complying with items 1 thru 3 and making appropriate logbook entry, return engine to service. Should the relief valve fail to lift off its seat within prescribed limits, the valve has malfunctioned. If so, refer to and comply with the Chart III overboost recommendations. Also, either reset or replace the pressure-relief valve.

CAUTION

ON ENGINES WHERE MANIFOLD PRESSURE IS THROTTLE-CONTROLLED BY THE PILOT, IN NO CASE ADVANCE THE THROTTLE BOOSTING MANIFOLD PRESSURE BEYOND RED-LINE TO DETERMINE IF THE ABSOLUTE PRESSURE RELIEF CONTROLLER IS FUNCTIONING. THIS IS AN EMERGENCY CONTROLLER. DELIBERATE MANIFOLD PRESSURE OVERBOOST MUST BE AVOIDED.

NOTE: Revision "G" revises text, adds engine models to Chart I, and adds rpm values to Chart II.