



SCHWEIZER SERVICE NOTICE

NOTICE NO. N-9.2
DATE 3 October 1977
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MANDATORY

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SUBJECT: INSPECTION - POSSIBLE RETIREMENT, MAIN ROTOR BLADE

MODELS AFFECTED: All 269A, 269A-1, 269B and TH-55A Helicopters

- TIME OF COMPLIANCE:
- A. Shall be accomplished as follows, for all 269 Series Helicopters equipped with main rotor blade part numbers 239A1125 all serial numbers, 269A1131 all serial numbers, or 269B1145 blades with serial number prior to -1314.
 - 1. Within 10 hours of operation on all main rotor blades having more than 200 hours total time in service, unless already accomplished.
 - 2. At each subsequent 400 hours of operation, following initial inspection, up to and including 1000 hours of blade time in service.
 - 3. At each subsequent 100 hours of operation on main rotor blades having more than 1000 hours time in service until blades are retired from service.
 - B. Shall be accomplished as follows, for all 269 Series Helicopters equipped with main rotor blade part numbers 269A1131-1 all serial numbers; 269B1145 blades with serial number -1314 or subsequent; 269B1145-1 all serial numbers; or 269B1145-25 all serial numbers:
 - 1. Within 25 hours of operation for main rotor blades having more than 1000 hours total time in service.

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2. At each subsequent 100 hours of operation, following initial inspection, until blades are retired from service.

PREFACE: The information given in this Service Information Notice lists Maintenance and Inspection procedures to ensure that main rotor blades are free of cracks and/or fatigue damage in the area of the root fittings. Inspection criteria are provided to help determine whether blade condition permits continued service, or necessitates retirement of blade from service and replacement.

Compliance with the provisions of this notice does not affect the provisions of Service Information Notice No. N-10.1, dated 14 December 1967.

Reference

269 Series - Basic HMI, Issued 1 April 1973; Revision No. 4, 15 Dec 1976
FAA Airworthiness Directive 64-28-2
FAA Airworthiness Directive 68-17-7
Hughes Service Information Notice No. N-10.1, dated 14 Dec 1967
Hughes Service Information Letter L-38.1, dated 15 July 1968
Hughes Service Information Notice N-46, dated 18 April 1968
Hughes Service Information Notice N-47, dated 18 April 1968

(1) Bar denotes portion of text superseded.

TOOLS AND EQUIPMENT

Glass - Magnifying 8 to 12 x	Lamp - Floor
Bolt (2) NAS 464P4A9	Solvent - Noncorrosive
Alternate bolts NAS 1104, MS20004, NAS 624, AN24	

PRELIMINARY PROCEDURE

- a. Using hacksaw, remove heads from two NAS bolts.
(See Figure 1)
- b. Using hacksaw, cut screwdriver slot on unthreaded end of bolt and deburr.
- c. Determine that compliance with requirements of Service Information Notices N-46 dated 18 April 1968, or N-47 dated 18 April 1968 has been accomplished.

NOTE

If not already accomplished, compliance with Service Information Notice N-46 or N-47 shall be performed in conjunction with the Maintenance and Inspection Procedure listed below.

MAINTENANCE AND INSPECTION PROCEDURE

- a. Remove three main rotor blades from rotorcraft
(See Reference).
- b. Inspect general condition of blades (Refer to HMI).

NOTE

When inspecting the areas of the blades defined in the following steps, use the magnifying glass. The inspection should be performed in direct sunlight or by using the floodlight. When using the floodlight, varying the position may be helpful. Any evidence of cracking is cause for retiring the blade from service. Cracked blades should be marked to prevent inadvertent return to service.

c. Using marking pen, number bolts securing root fitting from outboard to inboard.

d. Remove root fitting retaining bolts 3 and 5 from top of blade (See Figure 1).

WARNING

Serious damage can result if bolt washer combinations are improperly installed. Washers are dimensionally matched to each bolt and bolt/washer combination is matched to each hole.

e. Retain washers with bolt.

f. Install two fabricated studs through holes 3 and 5 into nutplate.

CAUTION

Studs are used to retain the nut plates on the inside of the blade. If studs are not installed, nut plates may be lost inside of blade.

g. Remove bolts 1, 2, 4 and 6; maintain bolt/washer integrity.

h. Remove root fitting and phenolic spacer from blade.

i. Clean blade, using soft cloth and suitable noncorrosive solvent.

j. Inspect for cracks, holes 1 through 6, paying particular attention to 1 and 2. (See Figure 2)

k. While performing inspection of holes, transfer nut plate retaining studs one at a time to previously inspected hole.

NOTE

While inspecting the blade holes for hairline cracks, examine the area at the forward and aft sides of the holes; and at both the skin surfaces and the inside of the hole itself, including the spar.

1. If blade/ blades are not cracked, reinstall root fitting and bolt/washer combinations in reverse order of removal, noting the following additions; torque bolts 55 to 60 inch-pound.
 1. Align root fitting and phenolic spacer on blade using nut plate retainer studs.
 2. Install bolt/washer combination carefully, to prevent binding or galling of holes, nut plates, and bolt.
 3. Torque bolts alternately; that is, 1 to 6, 2 to 5, 3 to 4.
- m. Install main rotor blades on rotor craft. (Refer to HMI)

FAA APPROVED

Stud Fabrication

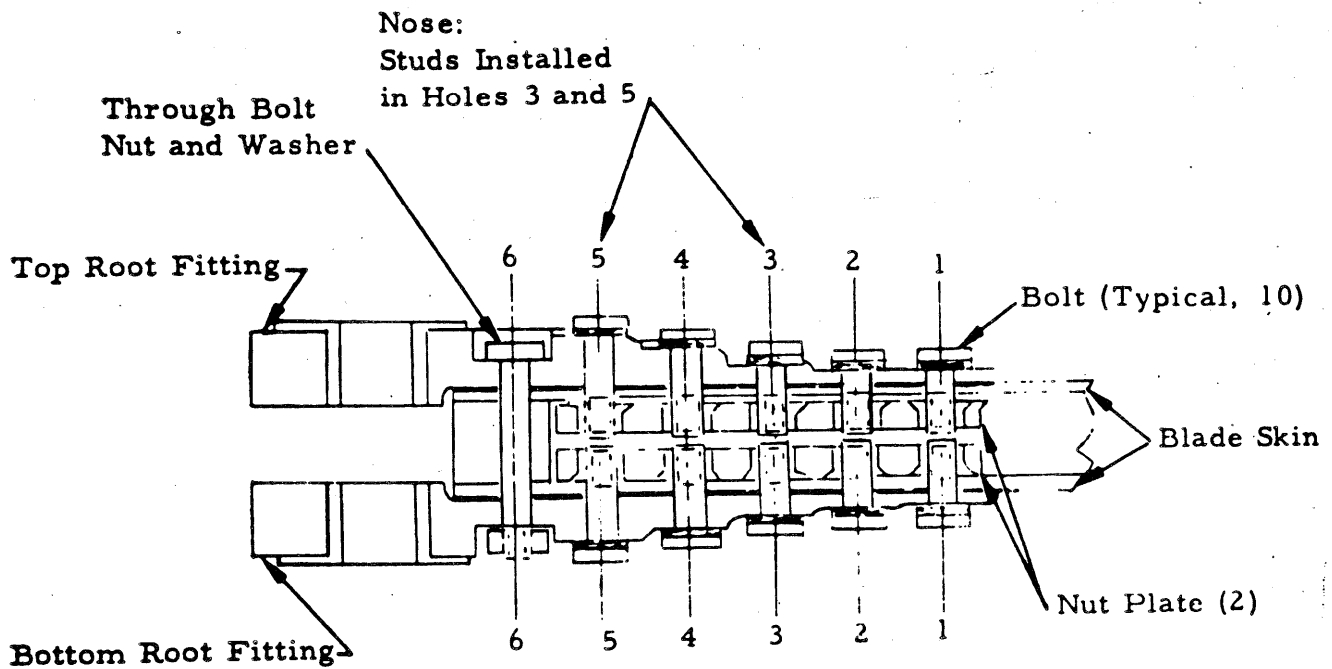
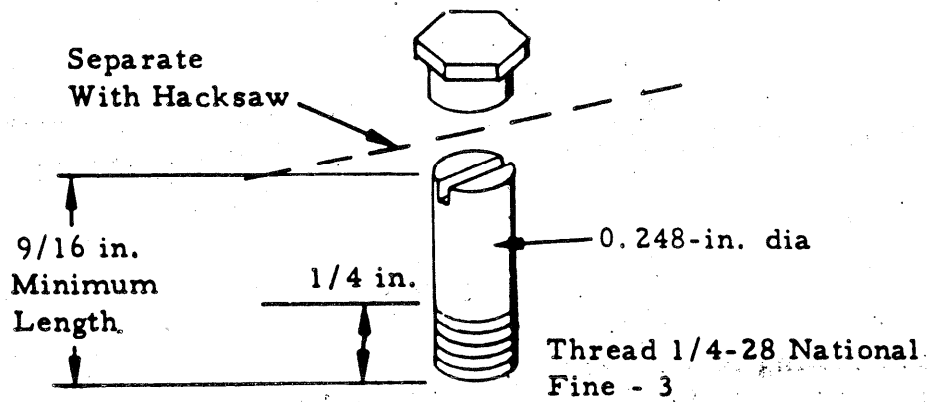


Figure 1. Main Rotor and Root Fitting Cross Section

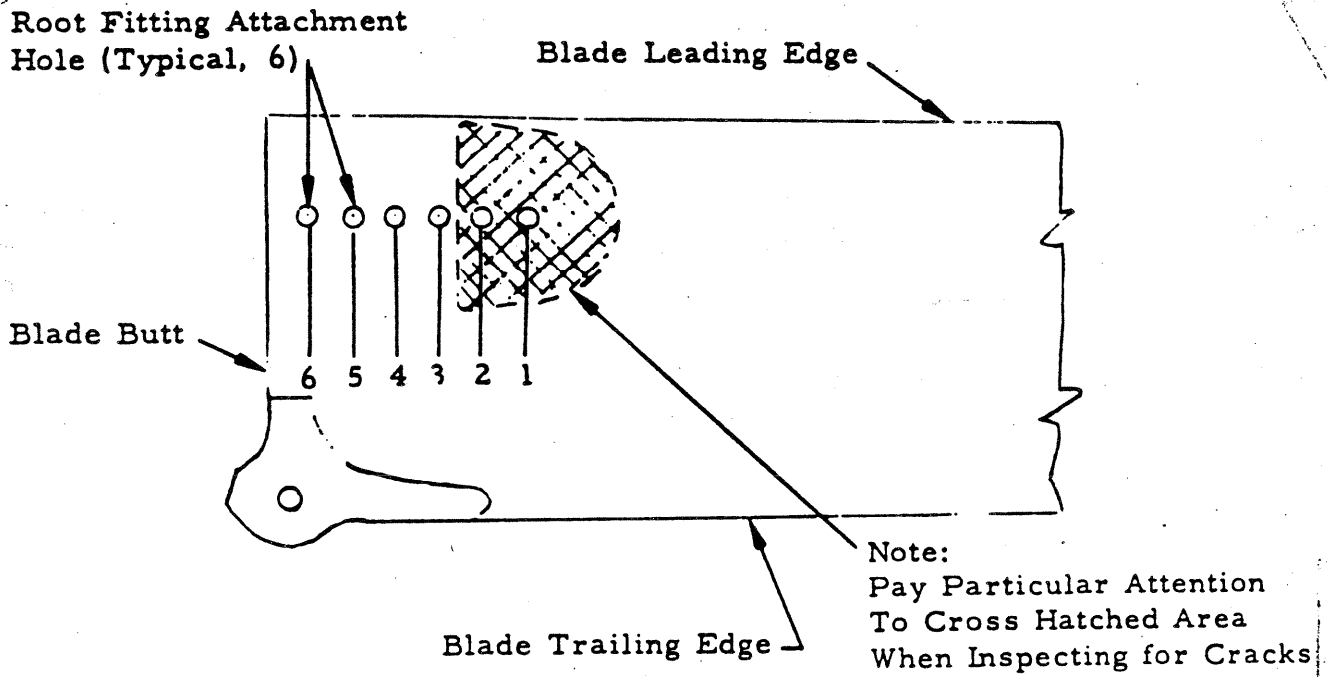


Figure 2. Main Rotor Blade Root Fitting Removed

